

Claims

1. Annular combustion chamber (4) for a gas turbine (1) wherein the annular combustion chamber (4) extends in an axial direction (A),
5 encloses a combustor (7), and has on its inside facing the combustor (7) a bearing structure (26) on which a lining element (10) secured to this lines the annular combustion chamber (4) characterized in that
on the rear side (13) facing away from the combustor (7) of two
10 edge areas (15) running in the axial direction (A) on the lining element (10) a plurality of interlocking means (11) are located which have a hook width (B), and in that the lining element (10) is secured to the corresponding bearing structure (26) such that in order to release the lining element (10) from the bearing
15 structure (26) the lining element (10) is moved by the extent of the hook width (B) of the interlocking means (11) in the axial direction (A).
2. Annular combustion chamber (4) according to Claim 1
20 characterized in that
a further plurality of interlocking means (11) are located as a central support (14) midway between two edge areas (15) of the lining element (10) running in the axial direction (A).
- 25 3. Annular combustion chamber (4) according to Claim 1 or 2
characterized in that
two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (A) have a spacing (L) which is identical to or greater than the hook width (B) of the
30 interlocking means (11).
4. Annular combustion chamber (4) according to one of the Claims 1 to 3
characterized in that

each interlocking means (11) has the identical hook width (B).

5. Annular combustion chamber (4) according to one of the Claims 1 to 4

5 characterized in that
two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (4) have a spacing (L) which is twice the hook width (B) of an interlocking means (11).

10 6. Annular combustion chamber (4) according to one of the Claims 1 to 4

characterized in that
two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (4) have a spacing (L)
15 which is three times the hook width (B) of an interlocking means (11).

7. Annular combustion chamber (4) according to Claim 5 or 6

characterized in that
20 each spacing (L) between two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (A) is identical.

8. Annular combustion chamber (4) according to one of the Claims 1 to 7

25 characterized in that
the lining element (10) has stiffening ribs (12) running in the circumferential direction (U) of the annular combustion chamber (4) on its rear side (13) facing away from the combustor (7).
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9. Annular combustion chamber (4) according to Claim 8

characterized in that
the stiffening rib (12) is distanced from the interlocking means (11).

10. Annular combustion chamber (4) according to one of the Claims 1
to 7
characterized in that
5 the interlocking means (11) are L- and/or T-shaped.
11. Gas turbine 1 with an annular combustion chamber (4) according to
one of the Claims 1 to 10.